

# **A displacement finite volume formulation for the static and dynamic analysis of shear deformable circular curved beams**

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## **Abstract**

In this paper, a finite volume formulation is proposed for static and in-plane vibration analysis of curved beams in which the axis extensibility, shear deformation and rotary inertia are considered. A curved cell with three degrees of freedom is used in discretization. The unknowns and their derivatives on cell faces are approximated either by assuming a linear variation of unknowns between the two consecutive computational points or by using the moving least squares technique (MLS). The proposed method is validated through a series of benchmark comparisons where its capability in accurate predictions without shear and membrane locking deficiencies is revealed.

*Keywords:* Curved beams; Finite volume; In-plane vibration; Membrane locking; Shear locking; Moving least squares

## **1. Introduction**

The numerical modeling of the curved beams is a challenging task due to the coexistence of three different load transfer mechanisms provided by the bending, shear and membrane flexibilities. The modeling of such coexistence of load carrying mechanisms may not be properly resolved

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which leads to deficiencies called as shear and membrane locking phenomena. These locking phenomena are seen in the finite element analysis using two-node curved element with linear shape functions. Researchers have proposed various schemes to alleviate such deficiencies including reduced integration [1], hybrid/mixed concepts [2], assumed displacement field [3], use of algebraic- trigonometric function [4], three- nodes curved beam element with trigonometric shape function [5], and use of discrete strain gap method [6]. Zhang and Di [7] presented accurate two-node finite element which was derived from the potential energy principle and the Hellinger–Reissner functional principle. They introduced the internal displacement parameters in developing a high-order related displacement- rotation interpolation field. Kim and Kim [8] used the nodeless degrees of freedom technique for developing an accurate, locking free hybrid-mixed  $C(0)$  curved beam element.

Unlike the straight beams, the governing equations of motion of curved beams are not only dependent on the rotation and radial displacement, but also on the coupled tangential displacement caused by the curvature of the structure. Wolf [9] employed straight elements to analyze the free vibration of elastic circular arches ignoring the effects of shear deformation but considering rotary inertia effects. Irie et al. [10] calculated the natural frequencies of in-plane vibrations of circular curved beams with uniform cross-section where the effects of rotary inertia and shear deformation are considered. Eisenberger and Efraim [11] studied the uniform circular beams using the Timoshenko beam theory, which includes the effects of rotary inertia, shear deformation, and the couplings of the radial and tangential displacements. Friedman and Kosmatka [12] utilized the Ritz method based on the trigonometric functions and derived the exact static stiffness matrix for the uniform circular curved beams. Yang et al. [13] have used a four-node Lagrangian type curved beam element and a polynomial equation of order 5 describing the tangential displacement, for modeling the uniform and non-uniform curved beams with variable curvatures.

All of the above-mentioned studies have been successfully applied to achieve the locking free elements with different accuracy levels. Most of the formulations deal with two-node or three-node beam elements. However, it should be mentioned that the higher accuracy of some of these elements is at the cost of inherent complex mathematical formulations.

In recent years, the finite volume (FV) technique which is a widespread and powerful technique in computational fluid dynamics, has been received a growing interest by the researchers for the solid mechanics analyzes. Elastic analysis of three dimensional solids [14], stress analysis of elasto-plastic solids [15], bending analysis of elastic plates [16-18], dynamic of solids[19], dynamic fluid-structure interaction [20], elasto-plastic analysis of plates [21] and active control of adaptive beams [22] are among the researches which expose the FV's capability in dealing with the variety of solid mechanics problems. FV formulations for the bending analysis of straight beams and plates have shown some advantages [17,18,21,22]: it is simple and transparent, it behaves well in the analysis of very thin to thick beams and plates and it predicts accurate results without using any adjustable parameter for the analysis of thin beams and plates.

Two types of finite volume technique are common: cell vertex finite volume [14,17,19,20] and cell centered finite volume [15-18,21-24].

In this paper, the finite volume method is applied for the static and dynamic vibration analysis of shear-flexible curved beams. Due to accounting for the transverse shear effects, the present formulation can be applicable for both thin and thick beams. For the finite volume analysis, first the beam is discretized into a mesh of curved elements known here as cells. For each cell, a computational point is considered at its center with three degrees of freedom. The internal forces acting on each cell are calculated and used in the balance equations of the cell. These equilibrium equations with equations expressing the boundary conditions provide a set of simultaneous linear equations which can be solved for the unknown calculations. For the calculation of the internal

forces, the interpolation of unknown variables is needed where two approaches are used for this purpose including the linear interpolation and interpolation based on the MLS technique. Although applying linear approach for the interpolation of unknowns in FV is common, however, using MLS technique in FV is rather new. Nevertheless, the MLS approximation technique has been known as a powerful technique in computer graphics [25], however, it now becomes widespread and is used in meshfree numerical methods for constructing shape functions.

To evaluate the proposed formulations, several numerical tests are studied where no shear and membrane lockings are observed in the analysis of Timoshenko curved beam models. Furthermore, the natural frequencies of some reference curved beams are calculated and the results are compared with the finite element and analytical results. Also, it is observed that using MLS increases the accuracy of the predictions.

This paper is presented in 6 sections. After section 1, curved beam formulation based on the cell centered finite volume technique is presented in section 2. Section 3 deals with the solution procedure for static and forced vibration analysis of curved beams. In section 4, free vibration analysis of curved beams is presented. Section 5 deals with some benchmark tests which are analyzed using the present formulation. Finally, in section 6, we draw the conclusions.

## **2. Formulation**

A portion of a circular curved beam is shown in Fig. 1(a), which is discretized to a number of 2-node curved elements along its centerline (Fig. 1(b)). Each element is considered as a control volume or cell. In the cell centered finite volume approach the computational points where the unknown variables are associated with, are coincided with the cell center. Three degrees of free-

dom are assigned to each computational point which are radial and tangential displacements and also the cross-sectional rotation shown by  $w$ ,  $u$  and  $\beta$ , respectively (Fig. 1(c)).

To obtain the finite volume based formulation, it is needed to determine the governing static and dynamic equilibrium equations of each individual cell. A control volume under a time dependant radial load is depicted in Fig. 2. The internal forces and moment acting on both sides of the cell and the inertia forces are shown in this figure where subscripts  $Ri$  and  $L$  denote the right and left sides of the cell, respectively. For the in-plane dynamic analysis of the beam, the governing equations of motion of a typical cell shown above are as follows

$$\sum M_c(t) = 0 \rightarrow M_I - \left[ M_{Ri}(t) - M_L(t) - (V_R(t) + V_L(t)) \times R \tan \frac{\Delta\theta}{2} \right] = m(t) + 2q_s(t)R^2 \text{tg} \frac{\Delta\theta}{2} (1 - \cos \frac{\Delta\theta}{2}) \quad (1)$$

$$\sum F_Y(t) = 0 \rightarrow F_{IY} - \left[ (V_{Ri}(t) - V_L(t)) \times \cos \left( \frac{\Delta\theta}{2} \right) + (N_{Ri}(t) + N_L(t)) \times \sin \frac{\Delta\theta}{2} \right] = 2q_Y(t)R \sin \frac{\Delta\theta}{2} \quad (2)$$

$$\sum F_S(t) = 0 \rightarrow F_{IS} - \left[ (N_{Ri}(t) - N_L(t)) \times \cos \frac{\Delta\theta}{2} - (V_{Ri}(t) + V_L(t)) \times \sin \frac{\Delta\theta}{2} \right] = 2q_S(t)R \sin \frac{\Delta\theta}{2} \quad (3)$$

where the terms with subscript  $I$  are the inertia related terms following the D'Alembert principle.  $M_I$  is the inertia moment associated with angular acceleration around the axis normal to the plane of motion passing through the cell center,  $F_{IS}$  and  $F_{IY}$  are also the inertia forces corresponding to the linear accelerations in  $S$  and  $Y$  directions respectively. It may be noted that the damping effects are neglected in the above equilibrium equations.  $\Delta\theta$  is the included angle corresponding to the cell and  $R$  is its initial radius of curvature.

The internal forces and moments appeared in the above equilibrium equations, acting on the cell faces, can be expressed in terms of the deformation related terms using the constitutive equations [26].

$$\begin{Bmatrix} M \\ V \\ N \end{Bmatrix}_j = \begin{bmatrix} EI & 0 & 0 \\ 0 & K_s GA & 0 \\ 0 & 0 & EA \end{bmatrix} \begin{Bmatrix} \frac{d\beta}{ds} \\ \frac{dw}{ds} + \beta + \frac{u}{R} \\ \frac{du}{ds} - \frac{w}{R} \end{Bmatrix}_j \quad (4)$$

where  $j$  stands for the right and left faces of the cell,  $EI$  is the flexural rigidity at the corresponding face,  $G$  is the shear modulus,  $A$  is the cross-sectional area of the corresponding face,  $K_s$  is the shear correction factor and  $R$  is the cell radius of curvature. Also,  $\beta$  is the cross-sectional rotation,  $u$  and  $w$  are displacements along the  $S$  and  $Y$  axis respectively. All these displacement components are corresponding to the face  $j$ .

The displacement components and also their gradients associated with both faces of the considered cell can be approximated in terms of displacement components measured at the centers of the considered cell and its neighboring cells. By using the interpolation techniques which are explained in the following section, approximating Eq. (4) and substituting the results into Eqs. (1) to (3) one obtains

$$\mathbf{M}_C \begin{Bmatrix} \ddot{\beta} \\ \ddot{w} \\ \ddot{u} \end{Bmatrix}_C + \mathbf{k}_C \begin{Bmatrix} \beta \\ \mathbf{w} \\ \mathbf{u} \end{Bmatrix}_C = \mathbf{P}_C \quad (5)$$

where  $\mathbf{P}_C$  contains the external loads and matrix  $\mathbf{M}_C$  is the diagonal mass matrix of the cell. The acceleration terms, i.e.  $\ddot{\beta}$ ,  $\ddot{w}$  and  $\ddot{u}$  are corresponding to the considered cell center, the dis-

placement terms which are gathered in the vectors,  $\boldsymbol{\beta}$ ,  $\mathbf{w}$  and  $\mathbf{u}$  are associated with the centers of considered cell and its neighboring cells. Matrix  $\mathbf{k}_C$  contains the coefficients relating the unknowns associated with the considered cell and its neighboring cells. Depending on the utilized interpolation schemes, these inter-cell relations can be established.

Diagonal mass matrix  $\mathbf{M}_C$  contains the whole mass of the cell which is lumped at the cell center as

$$\mathbf{M}_C = \begin{bmatrix} m_1 & 0 & 0 \\ 0 & m_2 & 0 \\ 0 & 0 & m_2 \end{bmatrix} \quad (6)$$

where  $m_1$  is the mass moment of inertia and  $m_2$  is the total mass of the cell.  $m_1$  and  $m_2$  are obtained using the following equations

$$m_1 = \rho I S_i \quad , \quad m_2 = \rho A S_i \quad (7)$$

in which  $A$ ,  $I$ ,  $S_i$ , and  $\rho$  are cross-sectional area, inertia moment, cell length and volumetric mass density of the cell respectively.

## 2.1 Interpolation

To calculate the unknown variables and their derivatives corresponding to the cell faces appeared in Eq. (4), two methods are used which are linear interpolation and MLS based interpolation methods. These methods are explained in the following parts.

### 2.1.1 Linear interpolation

A typical internal cell  $i$  and its neighboring cells are shown in Fig. 3. To calculate the unknowns and their required derivatives corresponding to the right face of the cell  $i$  which is a common

face of cell  $i$  and cell  $i+1$ , a linear variation of unknown variables is assumed between the centers of these two neighboring cells. Thus, the unknown variables and their first derivatives corresponding to that face can be calculated as [21,17,27]

$$\begin{aligned}
 \beta_r &= \beta_i + S_i^f \left( \frac{\beta_{i+1} - \beta_i}{S_{i+1}} \right) \\
 w_r &= w_i + S_i^f \left( \frac{w_{i+1} - w_i}{S_{i+1}} \right) \\
 u_r &= u_i + S_i^f \left( \frac{u_{i+1} - u_i}{S_{i+1}} \right)
 \end{aligned} \tag{8a-c}$$

$$\begin{aligned}
 \left( \frac{d\beta}{ds} \right)_r &= \frac{\beta_{i+1} - \beta_i}{S_{i+1}} \\
 \left( \frac{dw}{ds} \right)_r &= \frac{w_{i+1} - w_i}{S_{i+1}} \\
 \left( \frac{du}{ds} \right)_r &= \frac{u_{i+1} - u_i}{S_{i+1}}
 \end{aligned} \tag{9a-c}$$

Similar expressions can be written for the left face of the cell  $i$ . For a typical cell C, substitution of expressions (8) and (9) into constitutive Eq. (4) provides results that can be used in Eqs. (1), (2) and (3) which results in three equations in the form of Eq. (5).

Eq. (5) represents the relation of unknowns at the center of a cell to those at the centers of two neighboring cells. This procedure is applied for all the internal cells appeared in the model which provides three equations corresponding to each internal cell.

### 2.1.2 Higher order interpolation



The MLS method of interpolation is generally considered to be one of the best schemes to interpolate random data with a reasonable accuracy, because of its completeness, robustness and continuity [28]. According to the MLS, the distribution of a function  $u$  in  $\Omega$  can be approximated, over a number of scattered local points  $\{s_i\}$ , ( $i = 1, 2, \dots, n$ ), as,

$$u^h(s) = \mathbf{p}^T(s)\mathbf{a}(s) \quad \forall s \in \Omega \quad (10)$$

where  $\mathbf{p}^T(s) = [p_1(s), p_2(s), \dots, p_m(s)]$  is a monomial basis of order  $m$  and  $\mathbf{a}(s)$  is a vector containing coefficients, which are functions of the coordinates  $[s_1, s_2, s_3]$ , depending on the monomial basis.

In the present study the basis is chosen as

$$\mathbf{p}^T(s) = [1, s, s^2, s^3] \quad m = 4 \quad (11)$$

The coefficients included in vector  $a(s)$  are determined by minimizing a weighted discrete  $L_2$  norm, defined, as:

$$J(a(s)) = \sum_{I=1}^N W_I(s) [\mathbf{p}^T(s_I)a(s) - \hat{u}^I]^2 = [\mathbf{p}a(s) - \hat{u}^I]^T W_I(s) [\mathbf{p}a(s) - \hat{u}^I] \quad (12)$$

where  $s_I$  denotes the position vector of node  $I$ ;  $W_I(s)$  are the weighting functions and  $\hat{u}^I$  are the nodal parameters at  $s = s_I$ .  $N$  is the number of nodes in  $\Omega$  for which the weighting functions  $W_I(s) > 0$ . One may obtain the shape function as,

$$\mathbf{u}^h(s) = \boldsymbol{\phi}^T(s) \hat{\mathbf{u}} = \sum_{I=1}^N \phi^I(s) \hat{u}^I \quad (13)$$

where

$$\boldsymbol{\phi}^T(s) = \mathbf{p}^T(s) \mathbf{A}^{-1}(s) \mathbf{B}(s) \quad (14)$$

and its partial derivative is

$$\varphi_{,s}^I = \sum_{j=1}^m p_{j,s} (\mathbf{A}^{-1} \mathbf{B})_{jI} + p_j (\mathbf{A}_{,s}^{-1} \mathbf{B} + \mathbf{A}^{-1} \mathbf{B}_{,s}) \quad (15)$$

where matrices  $\mathbf{A}(s)$  and  $\mathbf{B}(s)$  are defined by

$$\mathbf{A}(s) = \mathbf{p}^T \mathbf{W} \mathbf{p} = \mathbf{B}(s) \mathbf{p} = \sum_{I=1}^N W_I(s) p(s_I) p^T(s_I) \quad (16)$$

$$\mathbf{B}(s) = \mathbf{p}^T \mathbf{W} = [w_1(s) p(s_1), w_2(s) p(s_2), \dots, w_N(s) p(s_N)] \quad (17)$$

The weighting function in Eq. (12) defines the range of influence of node I. Normally it has a compact support. In the present study  $N=4$  is considered and a quartic spline weighting function is used which is

$$\hat{W}(x - x_I) \equiv \hat{W}(\bar{d}) = \begin{cases} 1 - 6\bar{d}^2 + 8\bar{d}^3 - 3\bar{d}^4 & 0 \leq \bar{d} \leq 1 \\ 0 & \bar{d} > 1 \end{cases} \quad (18)$$

where

$$\bar{d} = \frac{d_I}{d_w} = \frac{|x - x_I|}{d_w} \quad (19)$$

in which  $d_I$  is the distance from node I to the sampling point and  $d_w$  is the size of the support domain for the weighting function.

Support domain  $\Omega_s$  is shown in Fig. 4. As can be seen in Fig. 4, for the internal cells of a beam discretized to the uniform cells, the size of  $d_w$  is 4 times of the length of the cells while for the end point cells the size of  $d_w$  is 2 times of the length of the cells. Eqs. (13), (14) and (15) can be used for approximating  $\beta$ ,  $w$ ,  $u$  and their derivatives corresponding to the right and left faces of each internal cell which are required in constitutive Eq. (4). Then the approximated constitutive Eq. (4) is substituted in equilibrium Eqs. (1) to (3) which results in the discretized equilibrium equations in the form of Eq. (5). The above approximating procedure is applied for all the internal cells which provides three equations corresponding to each internal cell.

## 2.2. Boundary conditions

To introduce the boundary conditions to the solution procedure, point cells are used at the two ends of the beam (Fig. 5). When displacement components are known at the end of the beam (displacement boundary condition), the incorporation of these known values in the solution procedure can be readily made. Depending on the interpolation schemes used, we have the following equations for imposing the displacement boundary conditions: in case of using the linear interpolations, we have:

$$\left\{ \begin{array}{c} \beta \\ w \\ u \end{array} \right\}_{P_b} = \left\{ \begin{array}{c} \beta^* \\ w^* \\ u^* \end{array} \right\} \quad (20)$$

and in case of using the MLS interpolation scheme we have:

$$\left\{ \begin{array}{l} \sum_{I=1}^N \phi^I(s) \hat{\beta}^I \\ \sum_{I=1}^N \phi^I(s) \hat{w}^I \\ \sum_{I=1}^N \phi^I(s) \hat{u}^I \end{array} \right\}_{P_b} = \left\{ \begin{array}{l} \beta^* \\ w^* \\ u^* \end{array} \right\} \quad (21)$$

where asterisked values denote the known applied values, and  $P_b$  indicates the point cell. When moment, shear and axial forces are known at the boundary (force boundary condition), we have

$$M_{P_b} = M^* \quad , \quad V_{P_b} = V^* \quad , \quad N_{P_b} = N^* \quad (22a-c)$$

where asterisked values denote the known applied values. Substituting constitutive equations with the approximated derivative expressions into the above equations give the appropriate equations. For linear interpolation method, we have:

$$\begin{aligned} \left[ \pm EI \left[ \frac{\beta_{P_b} - \beta_P}{S_b} \right] \right]_{P_b} &= M^* \\ \left[ K_s AG \left[ \frac{\pm (w_{P_b} - w_P)}{S_b} + \beta_{P_b} + \frac{u_{P_b}}{R} \right] \right]_{P_b} &= V^* \\ \left[ EA \left[ \frac{\pm (u_{P_b} - u_P)}{S_b} - \frac{w_{P_b}}{R} \right] \right]_{P_b} &= N^* \end{aligned} \quad (23a-c)$$

where the positive and negative signs are used for the right and left boundaries respectively. Parameters with subscript  $P_b$  are corresponding to the point-cell at the left or right boundaries and parameters with subscript  $P$  are corresponding to the adjacent cell next to the point cell. When MLS interpolation scheme is used in the finite volume formulation, the force boundary condi-

tions corresponding to a point cell like point cell at the left boundary can be expressed in the following form.

$$EI \left[ \phi_{1,s} \beta_{pb} + \phi_{2,s} \beta_2 + \phi_{3,s} \beta_3 + \phi_{4,s} \beta_4 \right]_{P_b} = M^* \quad (24)$$

$$K_s AG \left[ \begin{array}{l} \phi_{1,s} W_{pb} + \phi_{2,s} W_2 + \phi_{3,s} W_3 + \phi_{4,s} W_4 + \phi_1 \beta_{pb} + \phi_2 \beta_2 + \phi_3 \beta_3 + \phi_4 \beta_4 + \\ \frac{1}{R} (\phi_1 u_{pb} + \phi_2 u_2 + \phi_3 u_3 + \phi_4 u_4) \end{array} \right]_{P_b} = V^* \quad (25)$$

$$EA \left[ \phi_{1,s} u_{pb} + \phi_{2,s} u_2 + \phi_{3,s} u_3 + \phi_{4,s} u_4 - \frac{1}{R} (\phi_1 W_{pb} + \phi_2 W_2 + \phi_3 W_3 + \phi_4 W_4) \right]_{P_b} = N^* \quad (26)$$

where parameters with subscript 2,3,4 are associated with the centers of the first, second and third neighboring cells, respectively. By applying the boundary conditions for the two point cells, six equations are obtained.

### 3. Solution procedure for the discretized equilibrium equations

Equations associated with all the internal cells and two point cells provide a system of simultaneous linear equations which can be expressed in the matrix from:

$$\mathbf{M}\ddot{\mathbf{X}} + \mathbf{T}\mathbf{X} = \mathbf{P} \quad (27)$$

where  $\mathbf{T}$  contains the coefficients relating the unknown variables, and vector  $\mathbf{X}$  contains the unknown variables. Vector  $\mathbf{P}$  represents the known values on the boundaries and also the values that depend on the applied loads. Due to the sparse nature of  $\mathbf{T}$ , Eq. (27) can be solved by any appropriate solver technique for studying the bending behavior of the Timoshenko curved beam.

In case of existence of damping effects, the equation above can be re-expressed as

$$\mathbf{M}\ddot{\mathbf{X}} + \mathbf{C}\dot{\mathbf{X}} + \mathbf{K}\mathbf{X} = \mathbf{P} \quad (28)$$

where damping matrix  $\mathbf{C}$  can be approximated using Rayleigh damping [29] as follows

$$[\mathbf{C}] = \alpha[\mathbf{M}] + \beta[\mathbf{K}] \quad (29)$$

where  $\alpha$  and  $\beta$  are the proportional damping constants which are found using the following equation

$$\begin{bmatrix} \alpha \\ \beta \end{bmatrix} = \frac{2\xi}{\omega_m + \omega_n} \begin{Bmatrix} \omega_m \omega_n \\ 1 \end{Bmatrix} \quad (30)$$

where  $\xi$  is the damping ratio.  $\omega_m$  and  $\omega_n$  are frequencies of two different modes [29]. After determining the mass, stiffness and damping matrices, it is possible to state the dynamic equation of the control volume. The set of equations of the whole cells give a system of algebraic equations, which can be solved by applying an available solver technique. For solving the dynamic equilibrium Eq. (27) or (28), the Newmark method with a constant acceleration assumption is used [29].

#### 4. Free vibration

In order to study the harmonic vibrations of the beam, one can assume:

$$\begin{aligned}
\beta(s,t) &= \beta(s) \sin \omega t \\
w(s,t) &= w(s) \sin \omega t \\
u(s,t) &= u(s) \sin \omega t
\end{aligned} \tag{31a-c}$$

where  $\omega$  is the frequency of the natural vibration mode of the curved beam. By substituting above relations into Eq. (28) one can obtain the following eigenvalue equation

$$\mathbf{K}\bar{\mathbf{A}} = \lambda\mathbf{M}\bar{\mathbf{A}} \tag{32}$$

where  $\lambda$  contains the eigenvalues and  $\bar{\mathbf{A}}$  contains the natural modes of the vibrating beam.

## 5. Numerical results

To demonstrate the accuracy of the present formulations, some benchmark problems are studied and the results obtained are compared with the reference results reported in literature. At first, the performance of the present formulation is evaluated for the analysis of some straight beam tests. For this purpose, the present formulation which is corresponding to the curved beams is adjusted by using a large value for the radius of curvature,  $R$ . It is aimed to demonstrate the capability of the present formulation for the modeling of the straight beams under the static loads and also highlighting the shear locking free feature of the present method.

### 5.1 Uniformly loaded clamped-clamped straight beam

As a first evaluation test, the present formulation is used for the modeling of a straight uniformly loaded straight beam with the clamped–clamped boundaries. The material properties and geometry of the beam are:  $L= 3\text{m}$ ,  $h= 0.3\text{m}$  (depth),  $b= 0.01\text{m}$  (width),  $q= 1(\text{N/m})$

$E = 2.1\text{E}11 \text{ (N/m}^2\text{)}$ ,  $\nu= 0.3$ ,  $K_s= 5/6$ . The analytical prediction for the displacement at the middle of the beam span is calculated using the following equation given in Ref. [30]:

$$w\left(\frac{L}{2}\right) = \frac{qL^4}{384EI} \left[ 1 + .96(1 + \nu) \frac{h^2}{L^2} \right] \quad (33)$$

The beam is discretized by using an equally sized 2-node element and then analyzed by the present method. The error of the predictions of the proposed approach is obtained as follows:

$$Error = \left( \frac{w_{fv}}{w} - 1 \right) \times 100 \quad (34)$$

where  $w_{fv}$  is the predicted displacement and  $w$  is the value calculated using Eq. (33). For modeling the straight geometry of the beam, a large value is used for the parameter R in the discretized Eq. (27).

Fig. 6 illustrates how the errors diminish as the number of elements is increased by mesh refinement. The figure also shows that FVM with linear and MLS interpolation schemes predict the results which monotonically converge to the analytical solution. However, it may be noted that the results of FVM with MLS interpolation scheme converge to the corresponding analytical results faster than the FVM with linear interpolation technique.

Also, in Fig. 6 the results of the present method are compared with the results obtained by using ANSYS [31] software in which the equally sized element type of Beam 2D-elastic3 is used for the beam discretization. This element type is a 2-node element with six degrees of freedom which is able to model the shear effects. In order to have shear locking free behavior, the reduced integration technique [32,33] is used in the element formulation. As Fig. 6 shows the predictions of the present method with either linear or MLS interpolation schemes are more close to the analytical value of Eq. (34) than the ANSYS predictions. To investigate the aspect ratio, L/h, effect on the performance of the formulation, several aspect ratios, i.e. 5, 10, 100, 1000 are considered.



The error in the prediction of mid-span transverse displacement is calculated corresponding to each aspect ratio. As Fig. 7 demonstrates, the formulation is able to predict accurate transverse displacements of very thin to thick uniformly loaded clamped–clamped beams.

It is important to note that the shear locking phenomena is not observed in the analysis of thin Timoshenko beam models. It is well known that to eliminate the shear locking deficiency in the finite element analysis of Timoshenko beams, researchers have proposed techniques like reduced integration [32], selective integration [33], and some finite elements with rather complicated formulations [3,12,34].

### *5.2 Uniformly loaded cantilevered straight beam*

The material properties, geometry and load intensity of the beam are the same as the above test beam. FVM with both linear interpolation and MLS interpolation are used for the analysis of the beam. Again as we did in the above test, ANSYS with equally sized elements of Beam 2D-elastic3 is also used for the analysis of the beam. The error in the predictions of the tip displacement is calculated relative to the values given by [30]:

$$w_{tip} = \frac{qL^4}{8EI} \left[ 1 + 0.8(1 + \nu) \frac{h^2}{L^2} \right] \quad (35)$$

Fig. 8 illustrates the capability of the present formulation and ANSYS in different mesh densities in which FVM shows remarkable capability. Also to find out the performance of the FVM in dealing with the uniformly loaded cantilevered beam of different thicknesses, several aspect ratios, i.e. 5, 10, 100, 1000 representing thick to very thin beams, respectively, are considered. The error in the prediction of tip transverse displacement is calculated corresponding to each aspect

ratio. As shown in Fig. 9, the formulation is able to predict accurate tip transverse displacement of cantilevered beams in the range of thick to very thin beams.

### 5.3 Cantilevered straight beam with tip transverse load

In this test, a cantilevered beam with a tip transverse load is concerned. This beam has been used in Ref. [35] for evaluating the capability of the proposed finite elements in dealing with the shear locking deficiency. The material properties and geometry of the beam are,  $L= 4$ ,  $h= 0.554256$  (depth),  $b= 1$  (width),  $E = 2.6$  (N/m<sup>2</sup>),  $\nu= 0.3$ ,  $K_s= 0.85$ . The exact tip transverse displacement,  $w_T$ , is calculated using the following relation taken from Ref. [36]:

$$w_T = \frac{PL^3}{3EI} \left[ 1 + \frac{3EI}{KGAL^2} \right] \quad (36)$$

The tip displacement of the considered beam is obtained by using the finite volume method with both linear and MLS interpolation techniques which are shown in Table 1. As mentioned above, this beam has been already studied in Ref. [35] by using the Timoshenko finite element and a proposed higher order finite element formulation. In that work, Timoshenko finite elements have been formulated using both full integration and reduced integration, however, full integration has been used in the formulation of the proposed higher order finite element. In Table 1, the results of FVM and those from Ref. [35] are given in which ( $w_{TF}$ ), ( $w_{TR}$ ) and ( $w_{HOT}$ ) correspond to the results of Timoshenko finite element with full integration, Timoshenko finite element with reduced integration and higher order finite element with full integration, respectively. As can be seen, FVM with MLS interpolation predicts the analytical value by using just four elements. Al-

so FVM with linear interpolation produces results which are converging monotonically to the analytical value. While the FVM with either interpolation is able to analyze this test beam, the results shown in the second column of Table 1 indicate that the Timoshenko finite element formulation fails in accurate predictions due to shear locking phenomena. To remove the shear locking defect in the Timoshenko finite element formulation, the reduced integration technique has been used. By using this technique, Timoshenko finite element predictions converge to the analytical results, as shown in the third column of Table 1. Also the results obtained indicate that FVM with MLS interpolation converges faster than the higher order finite element formulation proposed by Ref. [35].

#### *5-4 A quarter circular cantilever ring*

As the first evaluation of the present formulation for the curved beam analysis, a quarter circular arch fixed at one end and free at the other end is studied. As shown in Fig. 10, the arch is subjected to a radial point force  $Q= 1\text{kN}$  at the free end and  $h=1\text{m}$ ,  $E = 5.6\text{E}6$  ( $\text{kN/m}^2$ ),  $G= 4\text{E}6$  ( $\text{kN/m}^2$ ). Two values for  $R$  are assumed as 10 and 50m which gives the slenderness ratios,  $R/h$ , equal to 10 and 50, respectively. The ring is divided into the equal sized curved elements or cells and analyzed by the present method. Also the ring is modeled by ANSYS using 2-node element of Beam 2D-elastic3, which is a straight element with six degrees of freedom. The displacement components at the free end are determined using both FV method and ANSYS, and then, the error of the predictions is calculated in comparison with the analytical values [3,7,34] calculated using Castigliano's energy theorem:

$$w = \frac{\pi QR^3}{4EI} + \frac{\pi QR}{4GAK_s} + \frac{\pi QR}{4EA} \quad (37)$$

$$u = -\frac{QR}{2EA} + \frac{QR^3}{2EI} - \frac{QR}{2GAK_s} \quad (38)$$

$$\beta = \frac{QR^2}{EI} \quad (39)$$

Figs. 11, 12 and 13 depict the errors in the predictions for both archs of R=10m and R=50m. It can be seen that both FV and ANSYS results are converging monotonically to the analytical values by increasing the element numbers. However, the FV predictions are more accurate than the ANSYS predictions when the same number of elements are used. It can be observed that even FV with linear interpolation predicts with more accuracy than the ANSYS.

#### *5-5 A circular arch under different static loads*

This test concerns an arc with included angle of 120 degree with different boundary conditions and under the different loading scenarios, in total of 7 cases. The arch's properties are: R= 4m, opening angle  $\theta=2\pi/3$  (length  $l=8\pi/3$ ), rectangular cross section with depth  $h= 0.6$  and width  $b= 0.4$ m, Young's modulus  $E= 30$ Gpa and poisson's ratio  $\nu=0.17$ . These archs have been already used for the evaluation of the formulations proposed by Litewka and Rakowski [4]. The arch is analyzed by the present method and also by ANSYS using straight element Beam 2D-elastic3.

The displacement components corresponding to the midpoint C are calculated and normalized

as:  $\bar{w}_c = \frac{w_c}{l}$ ,  $\bar{u}_c = \frac{u_c}{l}$ ,  $\bar{\beta}_c = \frac{\beta_c}{\theta}$ , where  $l$  is the arc length and  $\theta$  is the arc opening angle.

The calculated results and also the results of Ref. [4] are presented in Table 2.

It is observed that in all cases except the last one, the results of the FV with either linear or MLS interpolation techniques are found to be in good agreement with those reported in Ref. [4] and

the predictions of ANSYS. In case 7, the results of FV and ANSYS are in good agreement but far from the prediction reported in Ref. [4]. In that reference the formulation of a 2-node curved element has been developed by using the analytical shape functions of the trigonometric form. The element has been demonstrated to be shear and membrane locking free, however, does not perform properly in situations like case 7.

#### 5-6 Free vibration of a curved beam (pinned-pinned)

A uniform circular beam with the pinned-pinned boundary condition is considered as another test to demonstrate the capability of the present formulation. The beam has already been studied in Ref. [13] with the following material and geometrical properties:  $R/r = 15$ ;  $l/r = 23.56$ ;  $\varphi=90$ ;  $R=0.75\text{m}$ ;  $A = 4\text{m}^2$ ;  $I = 0.01\text{m}^4$ ,  $E= 70\text{GPa}$ ;  $k_s= 0.85$ ;  $k_sG/E = 0.3$ ;  $\rho= 2777\text{kg/m}^3$ .

where  $R$ ,  $r = \sqrt{\frac{I}{A}}$ ,  $l = R \varphi$ , and  $\gamma$  are the radius of curvature, the radius of gyration, length of the curved beam and the material density, respectively.

The non-dimensional natural frequencies ( $\gamma = \omega l^2 \sqrt{\frac{\rho A}{EI}}$ ) of the first 10 modes are calculated and presented in Table 3. It is observed that very close agreement exists between the present results and those published in Refs. [11,13,36]. The associated modal shapes are also presented in Fig. 14 where an excellent agreement is observed with the results given in Ref. [11]. It should be noted that in figures depicting the modal shapes, the horizontal axis represents the non-dimensional length parameter  $s/l$  in which,  $s$  is the tangential curvilinear coordinate and  $l$  is the total length of the beam.

### 5.7 Free vibration of a curved beam (clamped- clamped)

In this test, a uniform circular curved beam with the clamped boundaries is studied. The geometrical and material properties are the same as those used in Ref. [13]:  $R/r = 15.915$ ;  $l/r = 25$ ;  $\varphi = 90$ ;  $R = 0.6366\text{m}$ ;  $A = 1\text{m}^2$ ;  $I = 0.0016\text{m}^4$ ,  $E = 70\text{GPa}$ ;  $k_s = 0.85$ ;  $k_s G/E = 0.3$ ;  $\rho = 2777\text{ kg/m}^3$ .

The non-dimensional natural frequencies ( $\gamma = \omega l^2 \sqrt{\frac{\rho A}{EI}}$ ) of the first 10 modes are calculated and presented in Table 4. The close agreement between the results of the present study and the reference results can be observed. The associated modal shapes are also presented in Fig. 15 where an excellent agreement is observed with the results given by Ref. [11].

For analyzing the above last two tests, Ref. [11] has derived the finite element formulation using the shape functions which are the exact solutions of the differential equations of motion. Also, Ref. [13] applied an element of four nodes with 12 degrees of freedom. However, in the proposed formulation we use a curved cell of two points with three degrees of freedom in which even assuming a linear variation of unknowns between the computational points can provide accurate results.

### 5-8 forced vibration of a curved beam (clamped- clamped)

In this example, the forced vibration of a uniform circular curved beam with the clamped–clamped boundary is studied, Fig. 16. The geometrical and material properties are:  $R = 1\text{m}$ ,  $\varphi = 180$ ;  $b = .05\text{m}$ ;  $h = 0.2\text{m}$ ;  $E = 200\text{GPa}$ ;  $k_s = 0.85$ ;  $\nu = 0.3$ ;  $\rho = 7850\text{kg/m}^3$ .

A uniform distributed dynamic radial force of  $q_y = \sin \pi t$  is applied to this curved beam, and its radial vibration is studied. This beam is also analyzed by using ABAQUS [38] analysis tool where the element B21 is applied for the discretization. In the B21 element formulation the re-

duced integration technique is used for removing the locking deficiency. For the dynamic analysis of the beam by the present formulation, the FVM with linear interpolation is used. The Newmark approach [29] is utilized for solving the dynamic equilibrium equation in which the parameters of  $\gamma = 0.5$  and  $\beta = 0.25$  are used. In Fig. 17 the time history of radial displacement at the middle-span of the beam is shown and compared with the results predicted by ABAQUS. The close predictions of both approaches are evident. This comparison demonstrates the capability of the proposed formulation for this dynamic analysis.

#### *5-9 Forced vibration curved beam (pinned-pinned)*

The uniform circular curved beam with the pinned–pinned boundary condition is considered as the last test problem, Fig. 18. The beam has the following material and geometrical properties:

$$R= 2\text{m}; \varphi= 180; b= .05\text{m}; h= 0.2\text{m}; E= 200\text{GPa}; k_s= 0.85; \nu= 0.3; \rho = 7850\text{kg/m}^3$$

A concentrated dynamic force of  $q_y = \sin \pi t$  is applied at the middle-span of the beam. Similar to the previous example, FVM with linear interpolation and the ABAQUS software are used for the analysis of the beam. The time histories of the radial displacement of the middle-span of the beam are obtained using both methods and compared to each other. Fig. 19 shows the close predictions of both approaches.

## **6. Conclusions**

In this work the finite volume method presented in Refs. [23] and [27], has been extended for the modeling of the curved beams. The governing equations for the static and dynamic analyzes have been simply obtained by studying the equilibrium of each individual cell where the effects of the extensibility of the curved axis, the shear deformation and the rotary inertia are consid-

ered. For interpolation of unknowns, the linear approximation and MLS approximation schemes are used. To evaluate the proposed approach, several benchmark tests have been studied. It has been demonstrated that the proposed formulation is able to model both curved beams and straight beams where a large value for the radius of curvature is used for the straight beam models. In most test problems, MLS scheme presents more accurate results than the linear interpolation scheme. The convergence of the solutions to the exact results in the static test cases shows that the proposed simple formulation is free of shear and membrane locking deficiencies. Furthermore, the proposed approach has been applied to calculate the natural frequencies and forced vibration analysis of the several curved beams which have already been studied in literature. The comparison of the results obtained and the reference results also indicate the accuracy and effectiveness of the finite volume method for the vibration analysis of the curved beams. The attractive capabilities of the proposed approach in dealing with such rather challenging problem of curved beam models even using only 3 degrees of freedom for each cell (element) is promising.

## References

1. Stolarski, H. and Belytschko, T. "Membrane locking and reduced integration for curved elements". *J Appl Mech*, 49(1), pp. 172–176 (1982).
2. Reddy, B.D. and Volpi M.B. "Mixed finite element methods for the circular arch problem. *Comput Methods*", *Appl Mech Eng*, 97(1), pp. 125–145 (1992).
3. Raveendranath, P., Singh, G. and Pradhan, B. A two-noded locking-free shear flexible curved beam element. *Int J Numer Meth Eng*, 44(2), pp. 265–280 (1999).
4. Litewka, P. and Rakowski, J. "The exact thick arc finite element. *Comput Methods*", *Appl Mech Eng*, 68, pp. 369-379 (1998).
5. Saffari, H. and Tabatabaei, R. "A Finite Circular Arch Element Based on Trigonometric Shape Functions", *Math Probl Eng*,; Article ID. 78507, 19 pages, doi:10.1155/2007/78507 (2007).



6. Koschnick, F. and Bischoff, M. “The discrete strain gap method and membrane locking”, *Comput Methods Appl Mech Eng*, 194, pp. 2444–2463 (2005).
7. Zhang, C. and Di, S. “New accurate two-noded shear-flexible curved beam elements”, *Comput Mech*, 30, pp. 81–87 (2003).
8. Kim, J.G. and Kim, YY. “A new higher order hybrid mixed curved beam element”, *Int J Numer Meth Eng*, 43, pp. 925-940 (1998).
9. Wolf, J.A., “Natural frequencies of circular arches”, *J Struct Eng-ASCE*, 97, pp. 2337-50 (1971).
10. Irie, T., Yamada, G. and Tanaka, K. “Natural frequencies of in-plane vibrations of arcs” *J Appl Mech*, 50, pp. 449-52 (1983).
11. Eisenberger, M. and Efraim, E. “In-plane vibrations of shear deformable curved beams”, *Int J Numer Meth Eng*, 52, pp. 1221–1234 (2001).
12. Friedman, Z. and Kosmatka, J.B. “An accurate two-node finite element for shear deformable curved beams”, *Int J Numer Meth Eng*, 41, pp. 473– 498 (1998).
13. Yang, F., Sedaghati, R. and Esmailzadeh, E. “Free in-plane vibration of general curved beams using finite element method”, *J Sound Vib*, pp. 850–867 (1998).
14. Bailey, C. and Cross, M. “A finite volume procedure to solve elastic solid mechanics problems in three dimensions on an unstructured mesh”, *Int J Numer Meth Eng*, 38, pp. 1757–1776 (1995).
15. Demirdžić, I. and Martinović, D. “Finite volume method for thermo-elasto-plastic stress analysis”, *Comput Methods Appl Mech Eng*, 109, pp. 331-349 (1993).
16. Wheel, M.A. “A finite volume method for analysing the bending deformation of thick and thin plates”, *Comput Methods Appl Mech Eng*, 147, pp. 199-208 (1997).

17. Fallah, N. "A cell vertex and cell centred finite volume method for plate bending analysis" *Comput Methods Appl Mech Eng*, 193, pp. 3457-3470 (2004).
18. Fallah, N. "On the use of shape functions in the cell centered finite volume formulation for plate bending analysis based on Mindlin-Reissner plate theory", *Comput Struct*, 84, pp. 1664-1672 (2006).
19. Slone A.K., Bailey C. and Cross M. "Dynamic solid mechanics using finite volume methods", *Appl Math Modelling*, 2003; 27, pp 69-87 (2003).
20. Slone A.K., Pericleous K., Bailey C. and Cross M. "Dynamic fluid structure interaction using Finite Volume Unstructured Mesh procedures", *Comput Struct*, 80, pp 371-390 (2002).
21. Fallah, N. and Parandeh-Shahrestany, A. "A novel finite volume based formulation for the elasto-plastic analysis of plates", *Thin Wall Struct*, 77, pp. 153–164 (2014).
22. Fallah, N. and Ebrahimnejad, M. "Finite volume analysis of adaptive beams with piezoelectric sensors and actuators ", *Appl Math Model*, 38, pp. 722–737 (2014).
23. Fallah, N. "Finite volume method for determining the natural characteristics of structures", *J Eng Sci Tech*, 8(1), pp. 93-106 (2013).
24. Fallah, N. "A method for calculation of face gradients in two-dimensional, cell centred, finite volume formulation for stress analysis in solid problems", *Sci Iran*, 15, pp. 286-294 (2008).

25. Lancaster P. and Salkauskas K. "Surfaces generated by moving least squares methods", *Math Comput*, 37, pp. 141-158 (1981).
26. Day R.A. and Potts D.M. "Curved Mindlin beam and axi-symmetric shell elements- A new approach", *Int J Numer Meth Eng*, 30, pp. 1263 -1274 (1990).
27. Fallah N. and Hatami F. A displacement formulation based on finite volume method for analysis of Timoshenko beam, In: *Proceedings of the 7th International Conference in Civil Engineering: Iran*, May 8-10, (2006).
28. Liu G.R. and GU Y.T. "*An introduction to meshfree methods and their programming*", Springer press, Berlin (2005).
29. Chopra A.K. "*Dynamic of Structure, Theory and Applications to Earthquake Engineering*", Prentic-Hall, (1995).
30. Majkut L. "Free and forced vibration of Timoshenko beam described by single difference equation", *J Theor Appl Mech*; 47(1), pp. 193-210 (2009).
31. ANSYS, Swanson Analysis Systems Inc., *Ansys Reference Manual (version 12.1)*, (2009).
32. Hughes T.j.R., Cohen M. and Haroun M. "Reduced and selective integration techniques in the finite element analysis of plates". *Nucl Eng Des*, 46, pp. 203-222, (1978).
33. Bathe K.J. "*Finite element procedures*", Prentice-Hall, (1996).
34. Lee P.G. and Sin H.C. "Locking-free curved beam element based on curvature", *Int J Numer Meth Eng*, 37(6), pp. 989-1007, (1994).
35. Heyliger P.R. and Reddy J.N. "A higher order beam finite element for bending and vibration problems", *J Sound Vib*, 126(2), pp. 309-326, (1998).
36. Prathap G. and Bhashyam G.R. "Reduced integration and the shear-flexible beam element", *Int J Numer Meth Eng*, 18, pp.195-210 (1982).

37. Veletsos A. and Austin W. "Free vibration of arches flexible in shear", J Eng Mech-ASCE, 99, pp. 735–753 (1973).

38. ABAQUS, 6.12.1.

**Figure 1.** Discretization of a curved beam into the control volumes; (a) curvilinear coordinate system, (b) discretized beam, (c) degrees of freedom of the computational points.

**Figure 2.** A generic internal cell with the dynamic loads; (a) external loads, (b) internal loads.

**Figure 3.** A generic internal cell and its neighboring cells

**Figure 4.** Sub-domain for internal cell faces and point cells

**Figure 5.** Point cells at the boundaries

**Figure 6.** Error in the prediction of mid-span displacement of a uniformly loaded clamped-clamped straight beam ( $L/h=10$ ).

**Figure 7.** The effect of aspect ratio,  $L/h$ , on the prediction of mid-span displacement of a uniformly loaded clamped-clamped straight beam.

**Figure 8:** Error in the prediction of tip displacement of a uniformly loaded cantilevered straight beam ( $L/h=10$ ).

**Figure 9.** Error in the prediction of tip displacement of a uniformly loaded cantilevered straight beam with different  $L/h$ .

**Figure 10.** A quarter circular beam fixed at one end.

**Figure 11.** Error in the prediction of end point radial displacement.

**Figure 12.** Error in the prediction of end point tangential displacement.

**Figure 13.** Error in the prediction of end point sectional rotation.

**Figure 14.** Non-dimensional frequencies and vibration mode of a circular beam with uniform cross-section and pinned-pinned beam.

**Figure 15.** Non-dimensional frequencies and vibration mode of a circular beam with uniform cross-section and clamped-clamped boundary conditions.

**Figure 16.** Curved beam with the clamped-clamped boundary under dynamic load

**Figure 17.** Comparison of the time history of radial displacement of Middle-span of the arc between FVM and FEM method.

**Figure18.** Curved beam with the pinned–pinned boundary under the concentrated dynamic load.

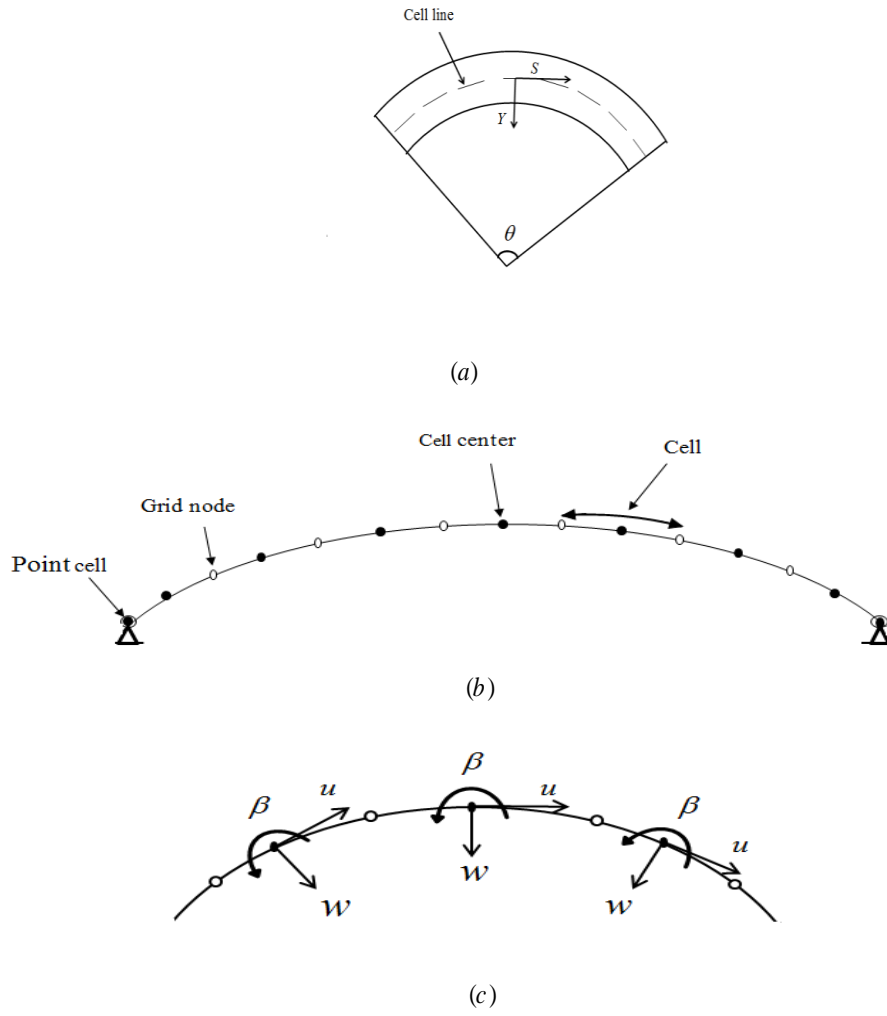
**Figure 19.** Comparison of the time history of the radial displacement of middle-span of the arc between FVM and FEM method.

**Table 1.**Comparison of displacements calculated by different approaches

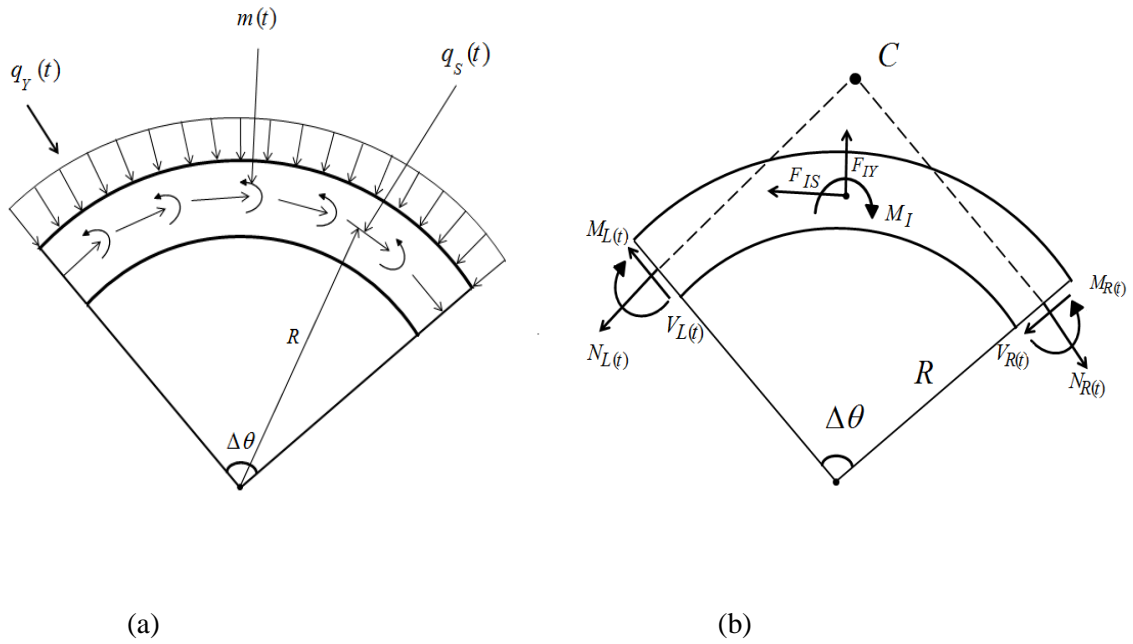
**Table 2.** Normalized displacements at the center of a circular beam under the different conditions.

**Table 3.** Non-dimensional frequencies  $\gamma$  of a uniform circular curved beam with the pinned–pinned boundary condition.

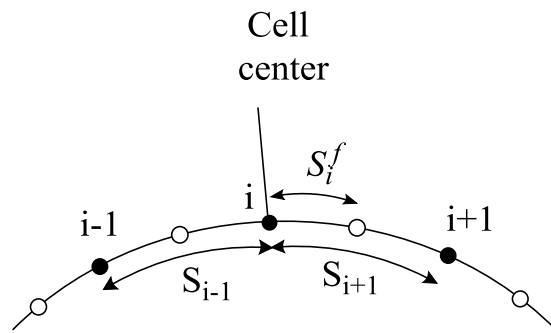
**Table 4.** Non-dimensional frequencies  $\gamma$  of a uniform circular curved beam with the clamped–clamped boundary condition.



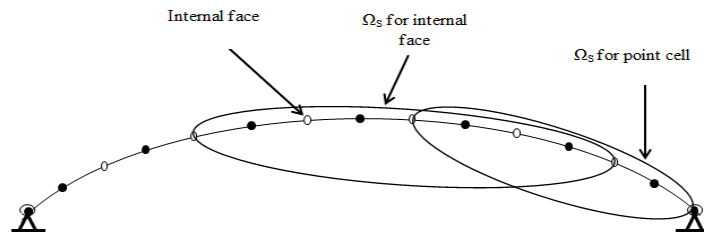
**Figure 1.**



**Figure 2.**



**Figure 3.**



**Figure 4.**



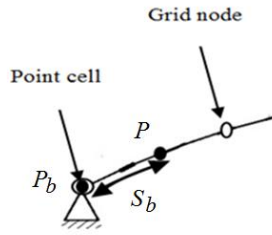


Figure 5.

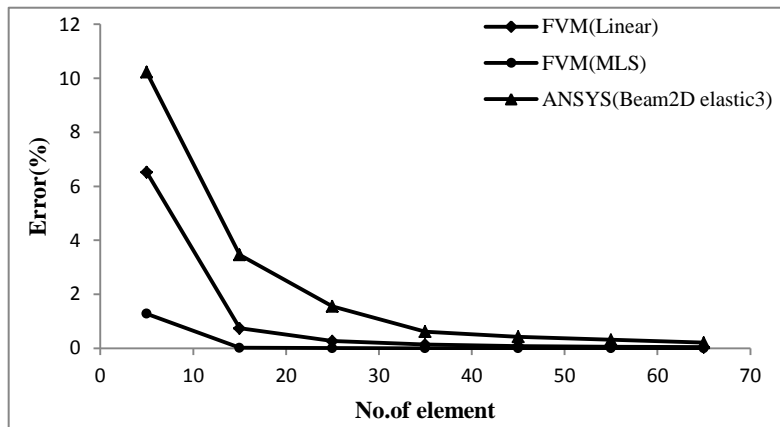


Figure 6.

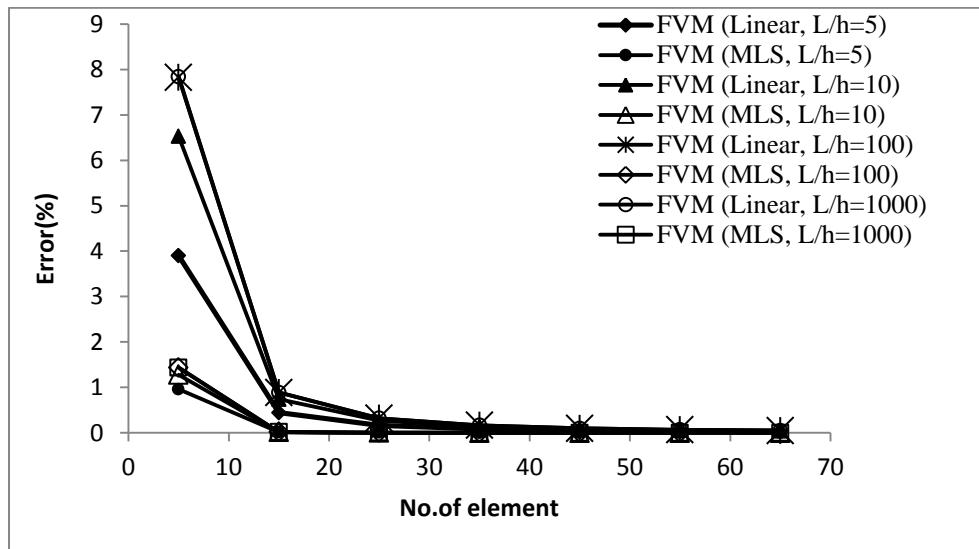


Figure 7.

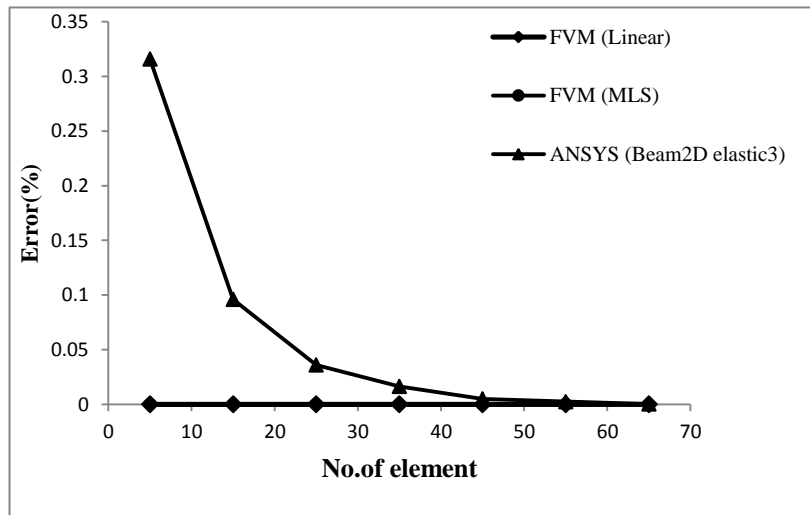


Figure 8.

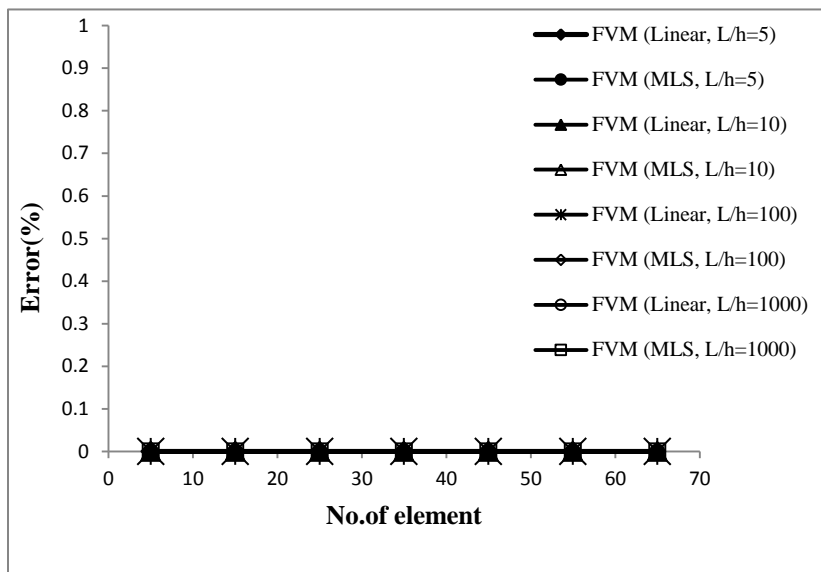


Figure 9.

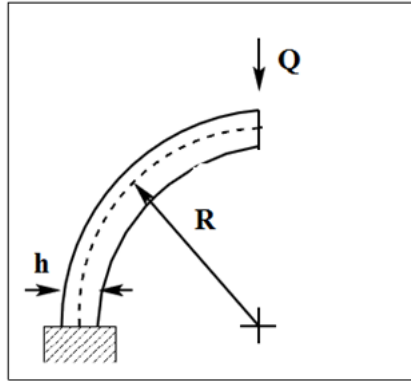


Figure 10.

Radial displacement (R=10, 50)

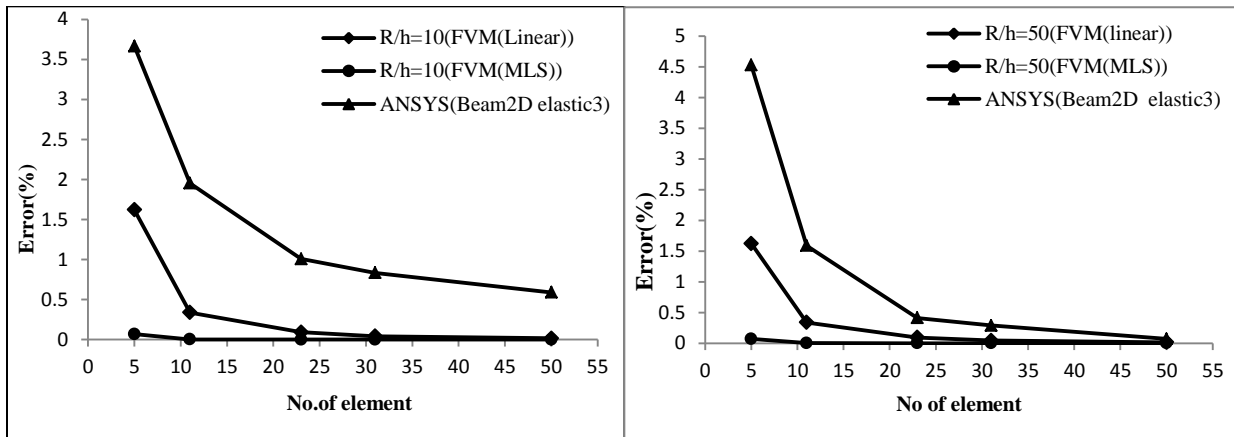


Figure 11.

Tangential displacement (R=10, 50)

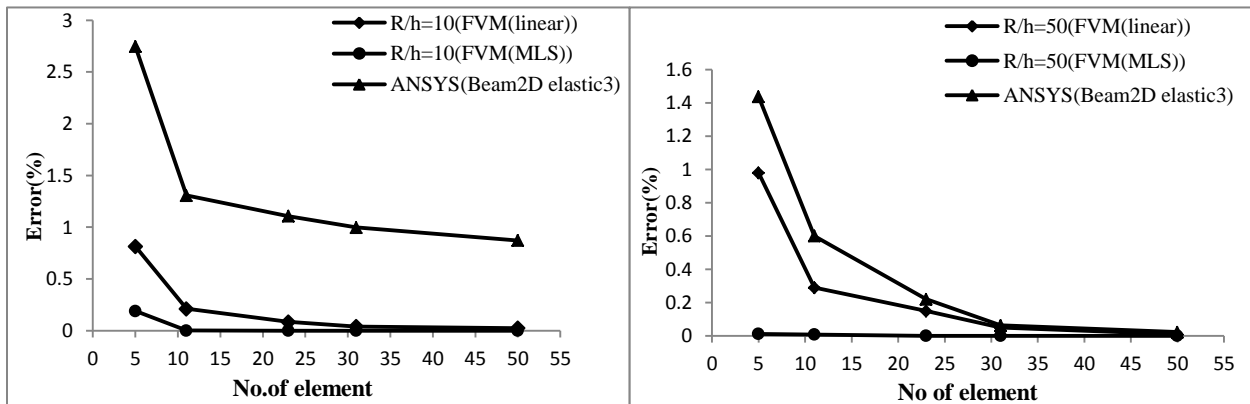


Figure 12.

Sectional rotation (R=10, 50)

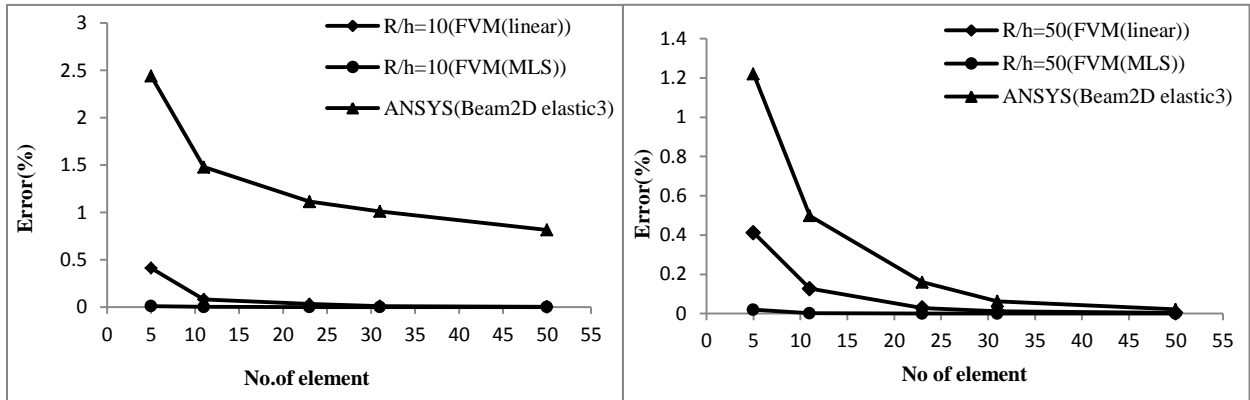
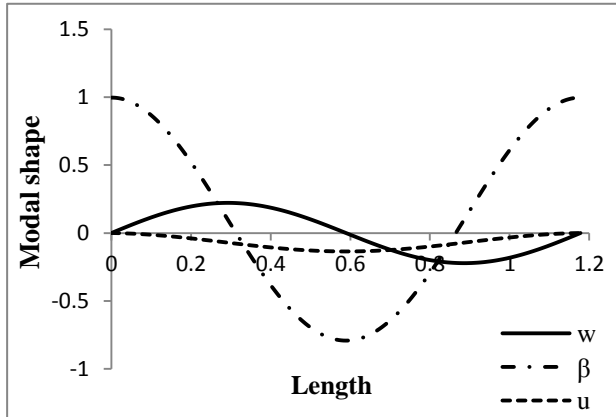
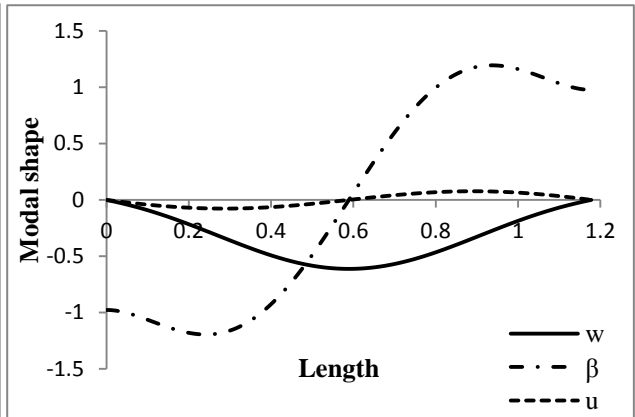


Figure 13.

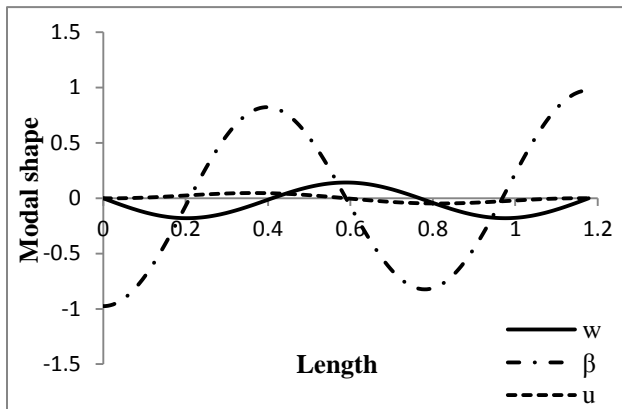
Vibration mode 1 ( $\gamma=29.261$ )



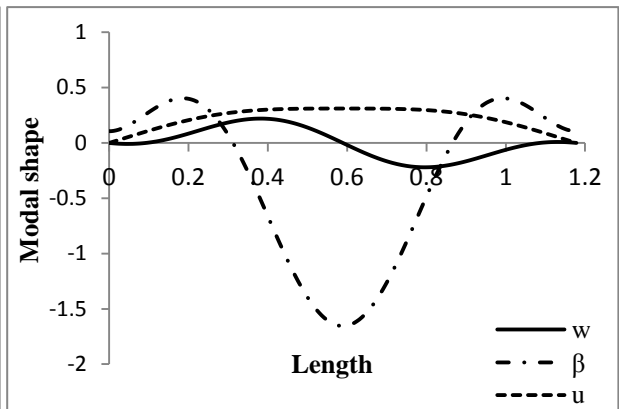
Vibration mode 2 ( $\gamma=33.297$ )



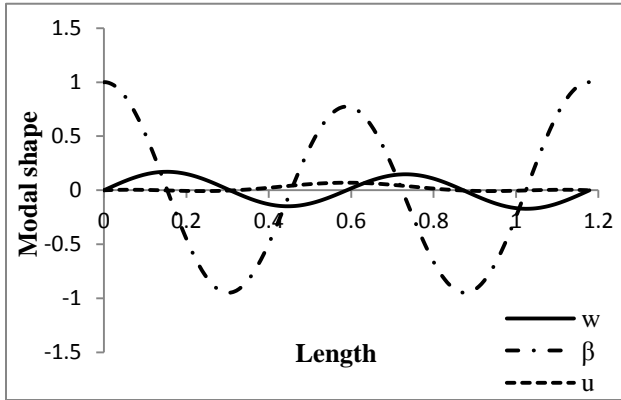
Vibration mode 3 ( $\gamma=67.101$ )



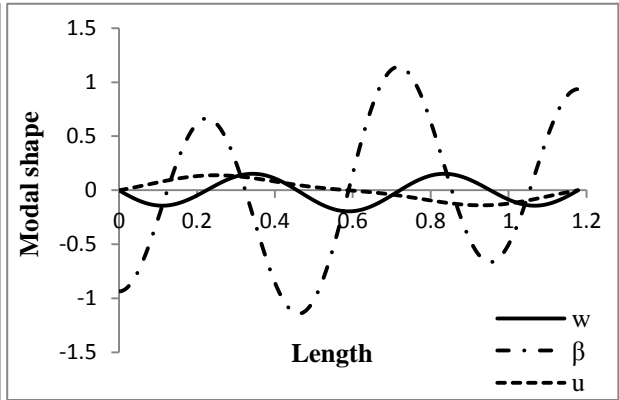
Vibration mode 4 ( $\gamma=79.957$ )



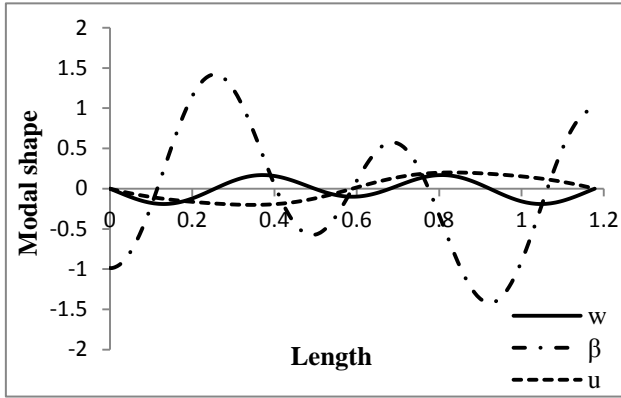
Vibration mode 5 ( $\gamma=107.825$ )



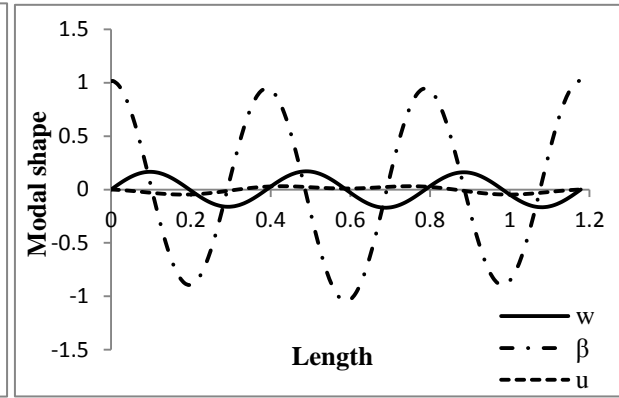
Vibration mode 6 ( $\gamma=143.59$ )



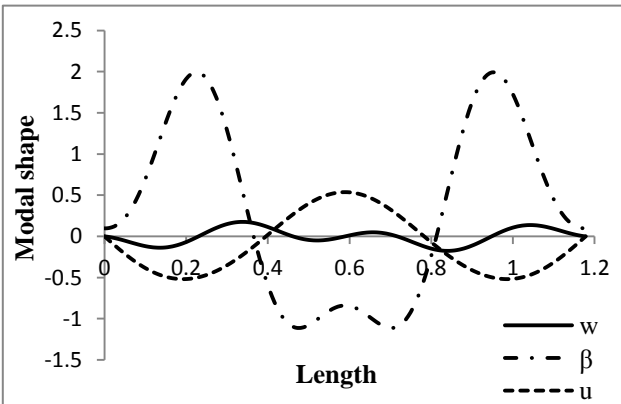
Vibration mode 7 ( $\gamma=156.636$ )



Vibration mode 8 ( $\gamma=190.439$ )



Vibration mode 9 ( $\gamma=225.320$ )



Vibration mode 10 ( $\gamma=234.473$ )

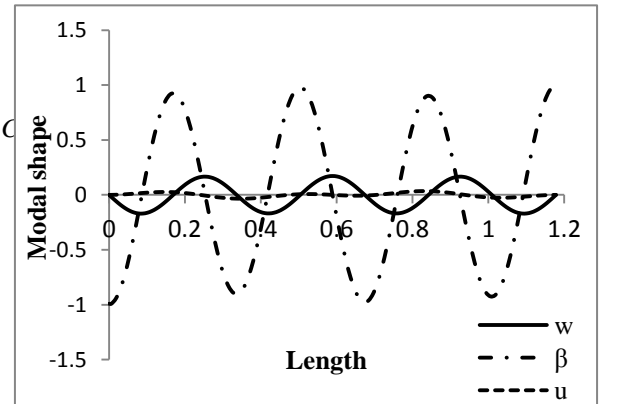
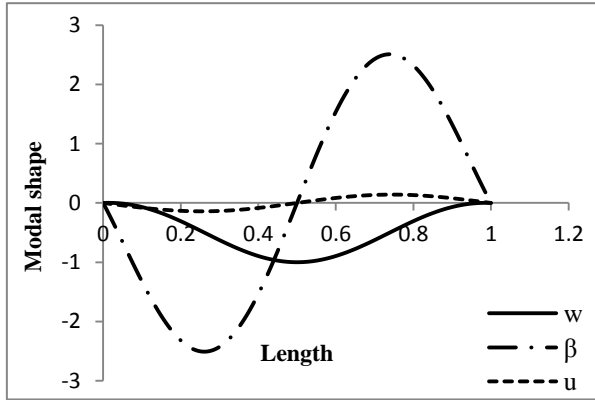
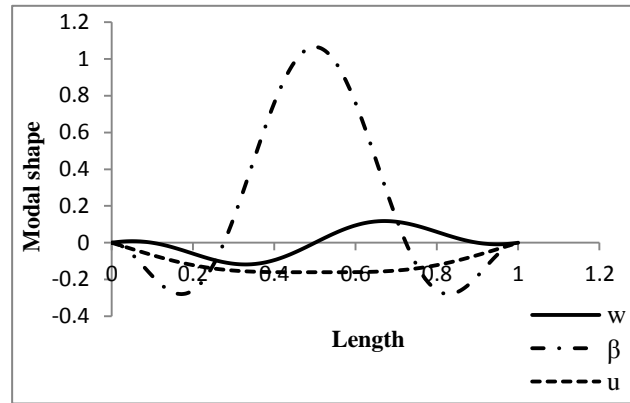
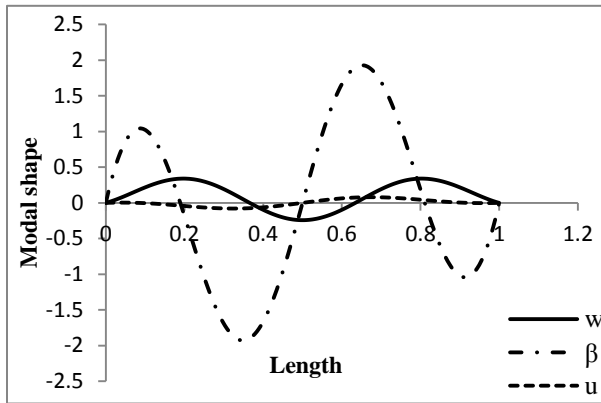
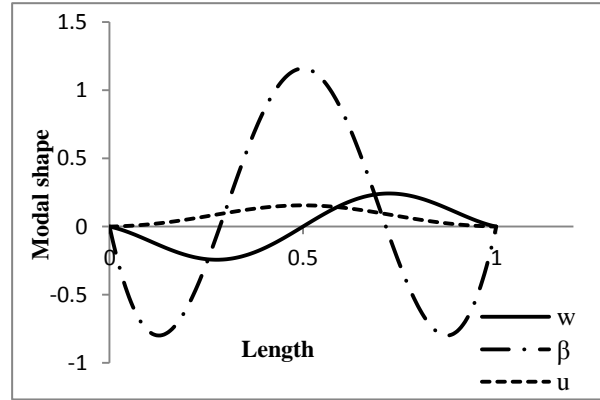


Figure14.

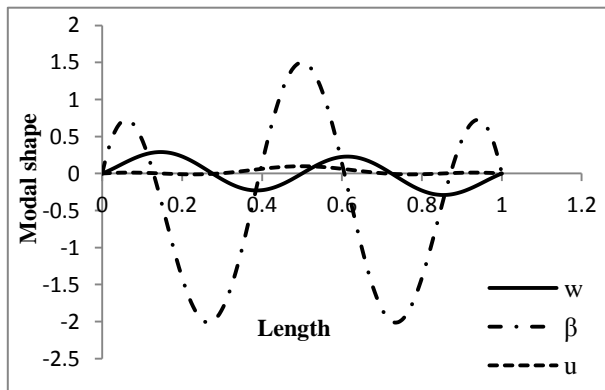
Vibration mode 1 ( $\gamma=37.705$ )



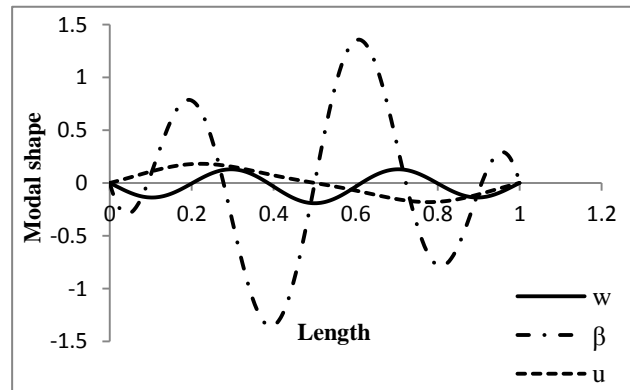
Vibration mode 2 ( $\gamma=42.268$ )



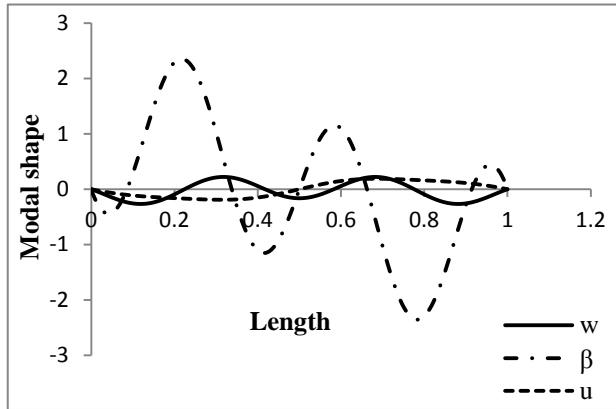
Vibration mode 5 ( $\gamma=122.311$ )



Vibration mode 6 ( $\gamma=154.95$ )



Vibration mode 7 ( $\gamma=168.208$ )



Vibration mode 8 ( $\gamma=204.479$ )

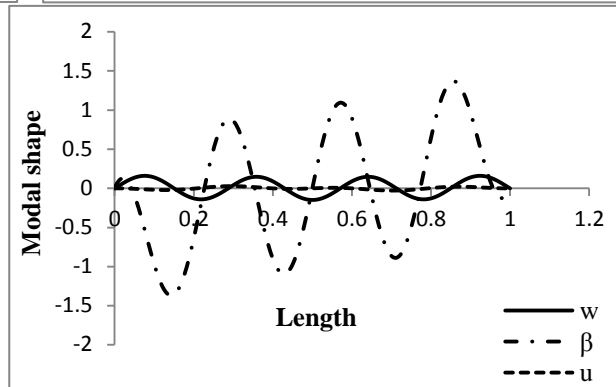
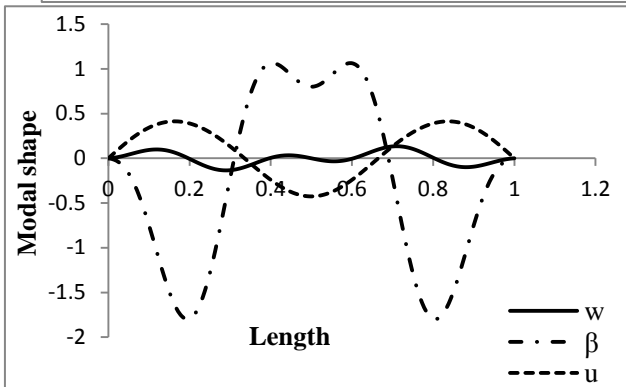
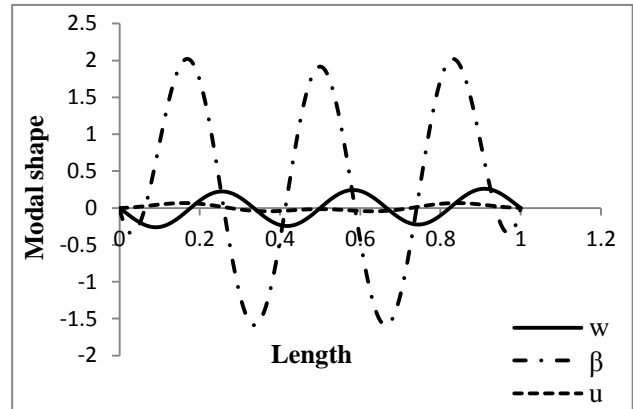


Figure 15.

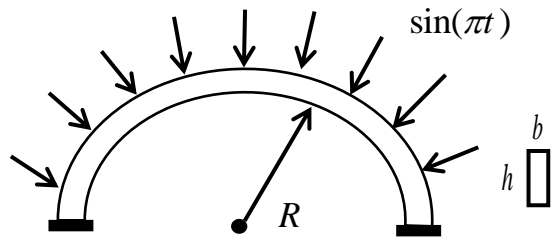


Figure 16.

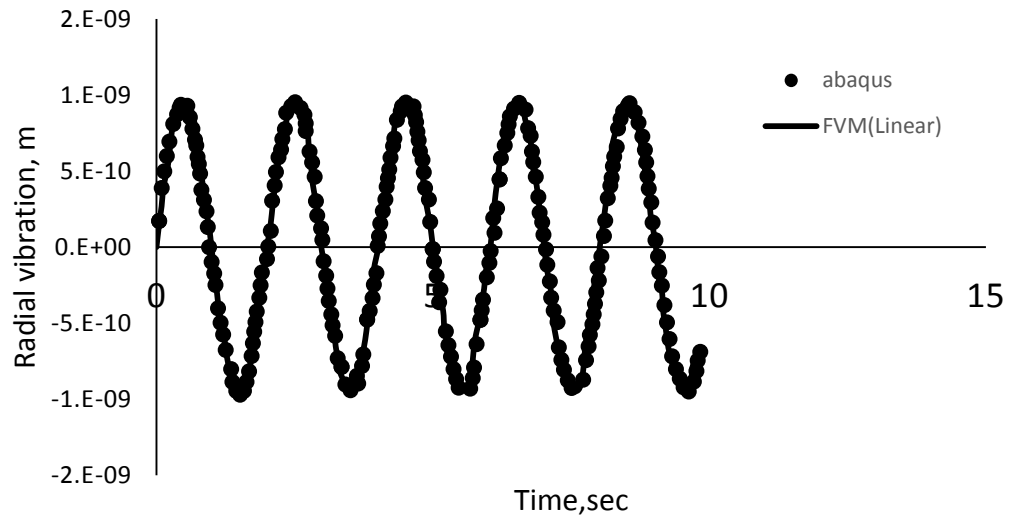


Figure 17.



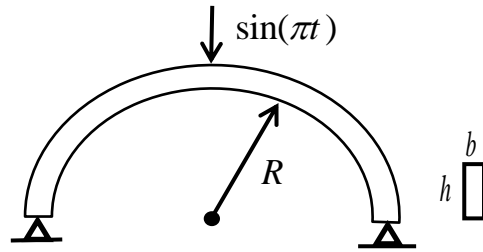


Figure 18.

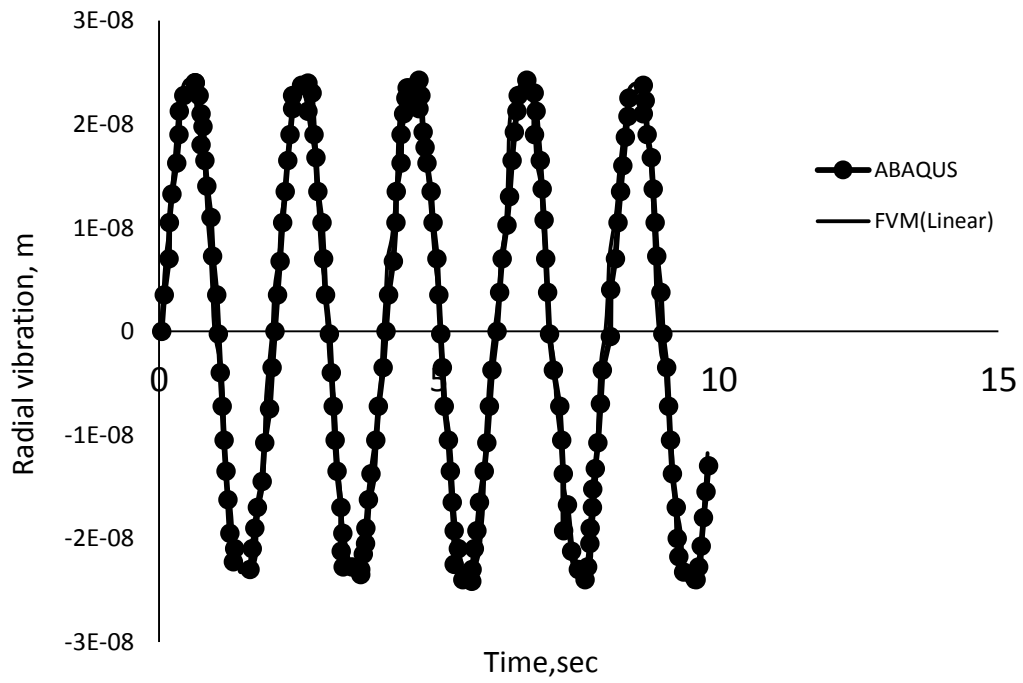
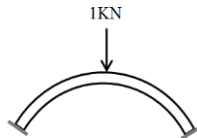
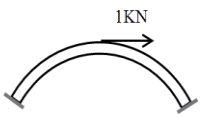
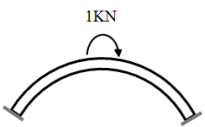
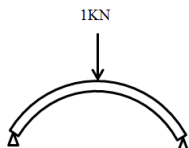
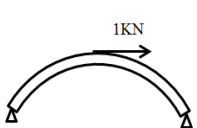
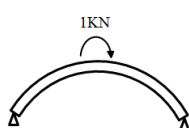
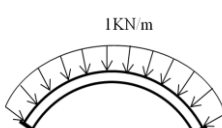


Figure 19.

**Table 1.**

No of Elements	$W_{TF}$ (FEM) [35]	$W_{TR}$ (FEM) [35]	$W_{HOT}$ (FEM) [35]	FVM (linear)	FVM (MLS)
2	111.6	550.6	553.3	550.6	-
3	202.9	570.7	573.1	570.7	-
4	284.2	577.7	579.9	577.7	586.8
5	349.0	581.0	582.9	581.0	586.8
6	398.4	582.8	584.4	582.8	586.8
7	435.6	584.2	585.3	583.8	586.8
8	463.5	584.6	585.8	584.5	586.8
9	485.0	585.2	586.1	585.0	586.8
(Exact)			586.8		

**Table 2.**

Case No	Arc	Normalized displacement	Litewka and Rakowski [4]	ANSYS	FVM (Linear)	FVM (MLS)
1		$\bar{w}_C (10^{-6})$	0.2488	0.2454	0.2485	0.2485
		$\bar{u}_C (10^{-6})$	0.0000	0.0000	0.0000	0.0000
		$\bar{\beta}_C (10^{-6})$	0.0000	0.0000	0.0000	0.0000
2		$\bar{w}_C (10^{-6})$	0.0000	0.0000	0.0000	0.0000
		$\bar{u}_C (10^{-6})$	0.1252	0.1202	0.1251	0.1251
		$\bar{\beta}_C (10^{-6})$	0.3796	0.374	0.3795	0.3794
3		$\bar{w}_C (10^{-6})$	0.0000	0.0000	0.0000	0.0000
		$\bar{u}_C (10^{-6})$	0.0949	0.0936	0.0948	0.0949
		$\bar{\beta}_C (10^{-6})$	1.0824	1.0695	1.0819	1.0749
4		$\bar{w}_C (10^{-6})$	0.3047	0.2715	0.2806	0.2807
		$\bar{u}_C (10^{-6})$	0.0000	0.0000	0.0000	0.0000
		$\bar{\beta}_C (10^{-6})$	0.0000	0.0000	0.0000	0.0000
5		$\bar{w}_C (10^{-6})$	0.0000	0.0000	0.0000	0.0000
		$\bar{u}_C (10^{-6})$	0.2884	0.2833	0.2881	0.2883
		$\bar{\beta}_C (10^{-6})$	0.8064	0.8021	0.8064	0.8060
6		$\bar{w}_C (10^{-6})$	0.0000	0.0000	0.0000	0.0000
		$\bar{u}_C (10^{-6})$	0.2016	0.2014	0.2015	0.2016
		$\bar{\beta}_C (10^{-6})$	1.3613	1.3537	1.3612	1.3540
7		$\bar{w}_C (10^{-6})$	0.118	0.475	0.4722	0.4721
		$\bar{u}_C (10^{-6})$	0.0000	0.0000	0.0000	0.0000
		$\bar{\beta}_C (10^{-6})$	0.0000	0.0000	0.0000	0.0000

**Table 3.**

Mode	FVM (Linear)	FVM (MLS)	Eisenberger and Efraim[11]	Yanga et al[13]	Veletsos and Austin [37]
1	29.261	29.284	29.28	29.306	29.61
2	33.297	33.301	33.305	33.243	33.01
3	67.101	66.120	67.124	67.123	67.24
4	79.957	79.958	79.971	79.95	79.6
5	107.825	107.838	107.851	107.844	107.7
6	143.619	143.596	143.618	143.679	144.5
7	156.636	156.640	156.666	156.629	155.2
8	190.439	190.447	190.477	190.596	191.3
9	225.320	225.322	225.361	225.349	223.7
10	234.473	234.483	234.524	234.809	235.3

**Table 4.**

Mode	FVM (Linear)	FVM (MLS)	Eisenberger and Efraim [11]	Yanga et al [13]	Veletsos and Austin [37]
1	37.705	36.704	36.703	36.65	36.81
2	42.268	42.267	42.264	42.289	42.44
3	82.239	82.237	82.233	82.228	82.5
4	84.494	84.494	84.491	84.471	84.3
5	122.311	122.310	122.305	122.298	122.5
6	154.949	154.949	154.945	154.998	155.1
7	168.208	168.208	168.203	168.174	167.7
8	204.479	204.478	204.472	204.599	-
9	238.998	238.997	238.992	238.973	-
10	248.019	248.016	249.011	249.320	249.6

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